

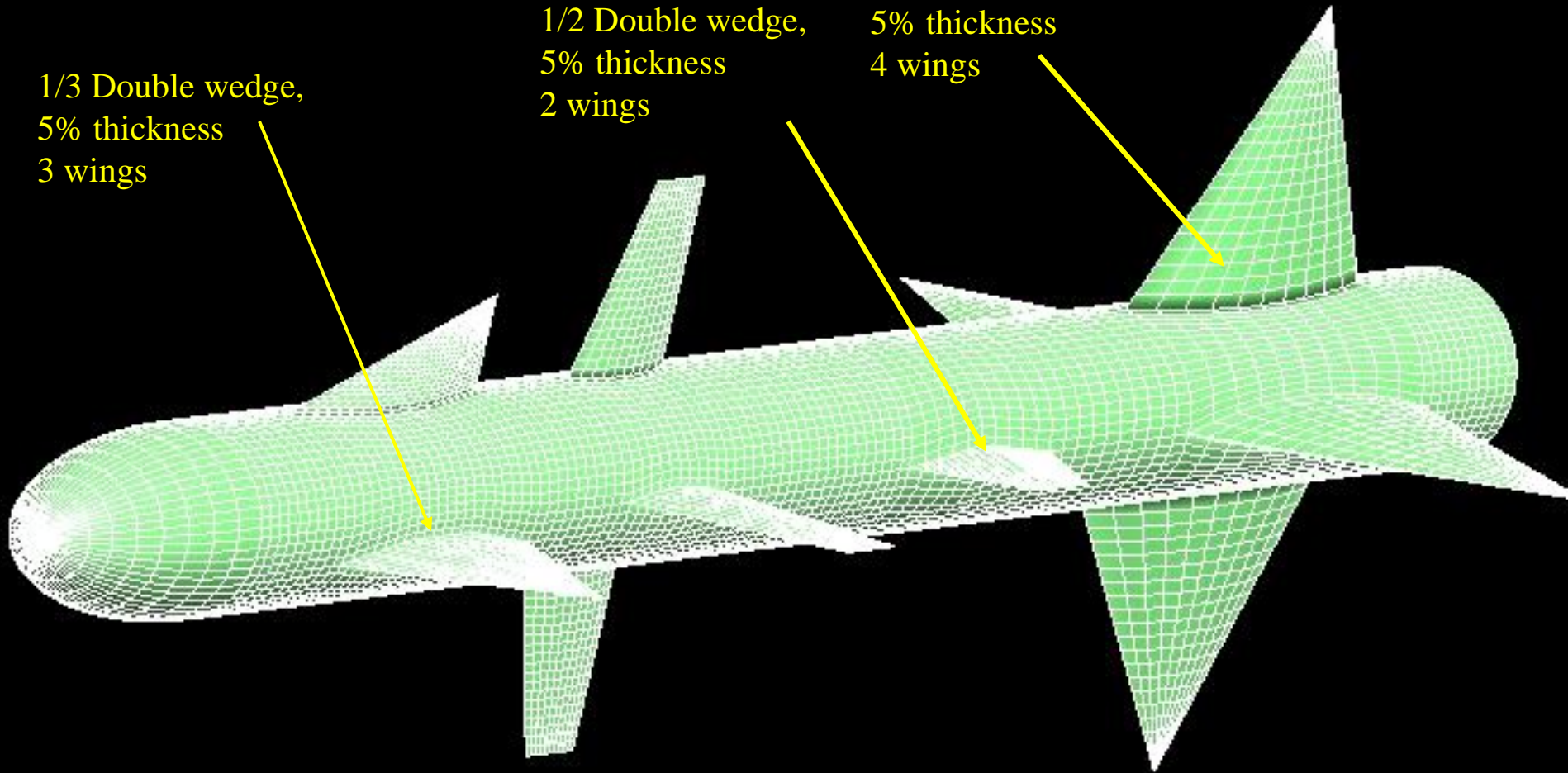
Flow Past Canard-Wing-Tail-Body

- Mach 1.2 flow past a canard-wing-tail-wing-body at 10 degrees angle of attack
- B

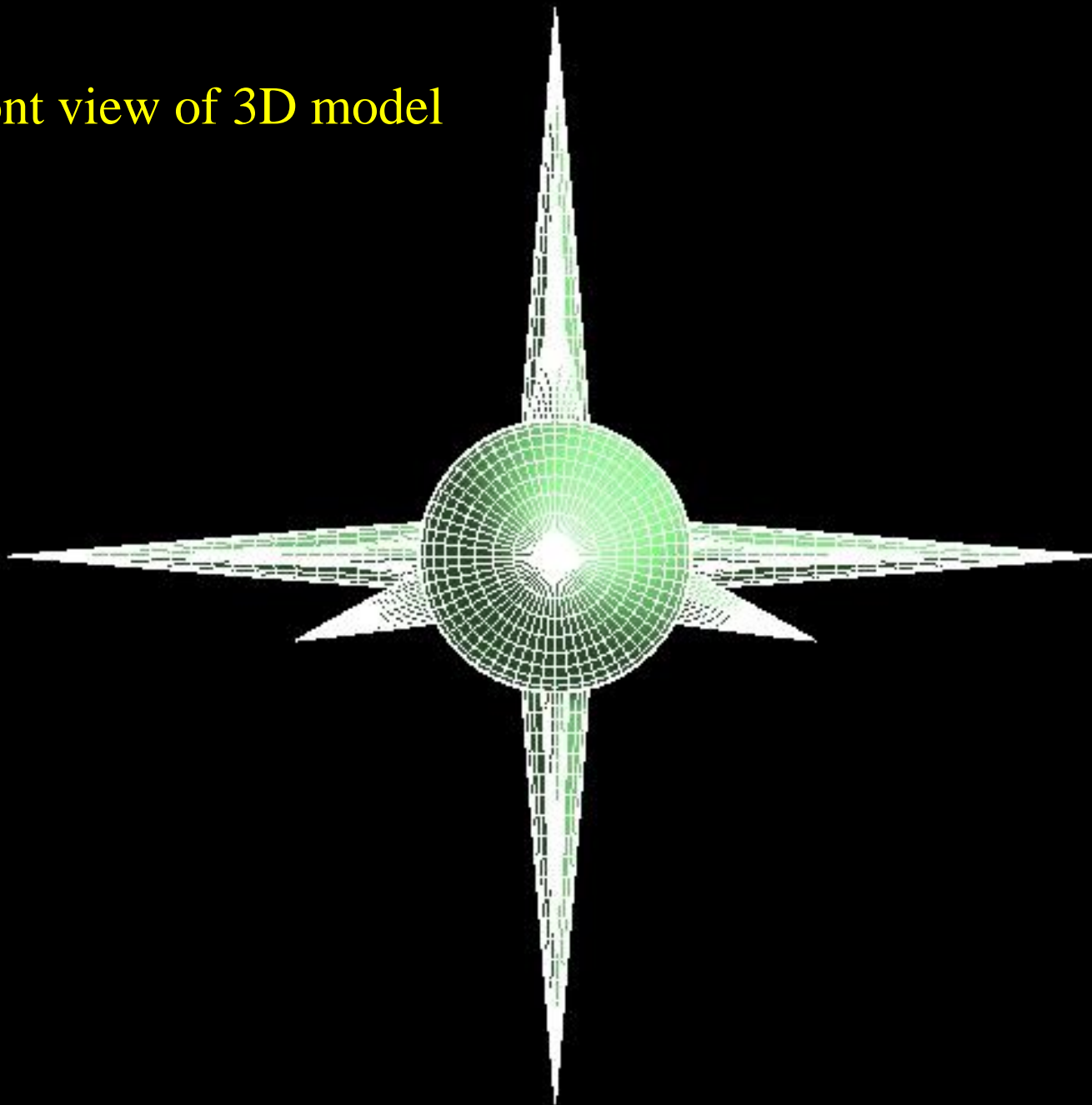
1/3 Double wedge,
5% thickness
3 wings

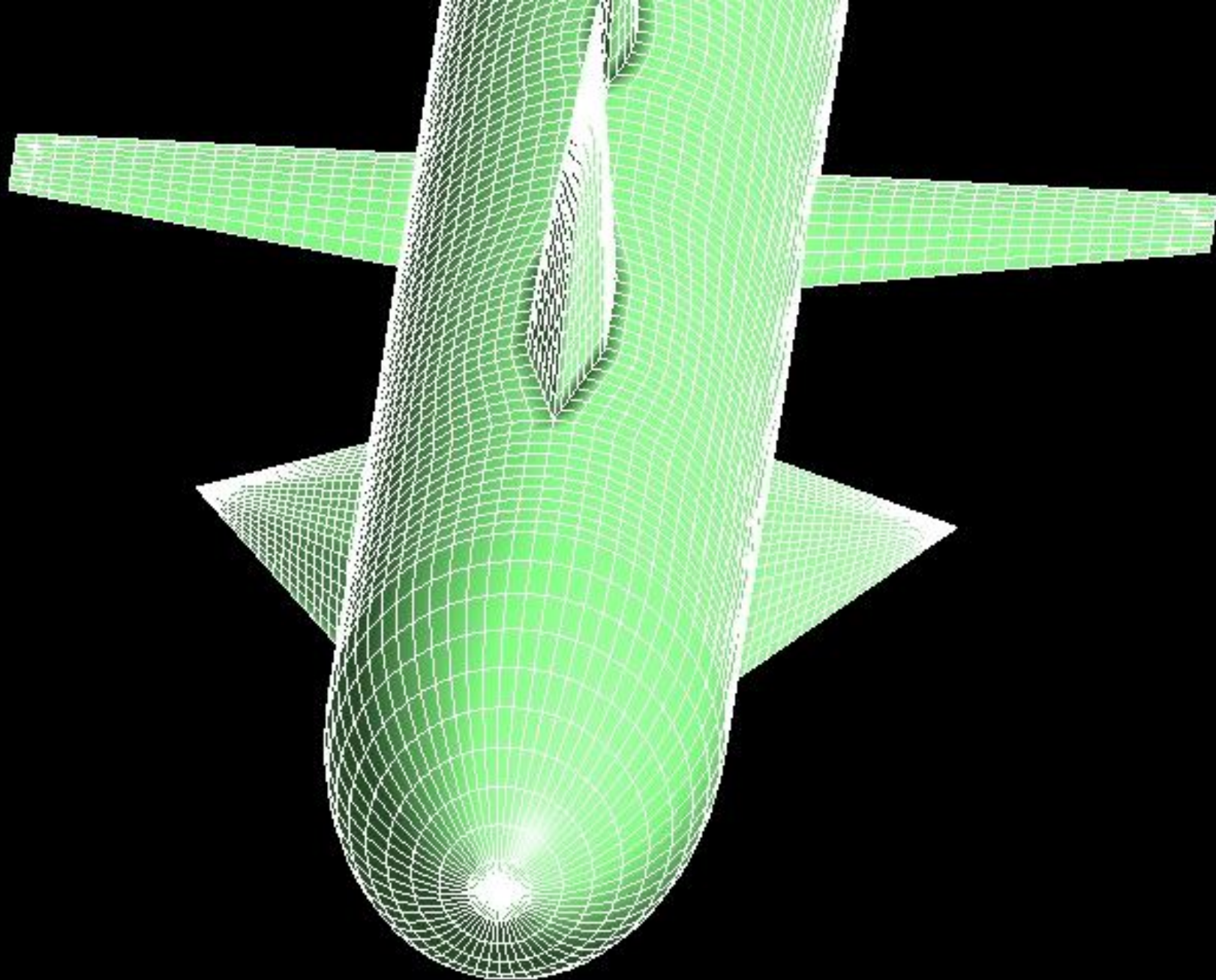
1/2 Double wedge,
5% thickness
2 wings

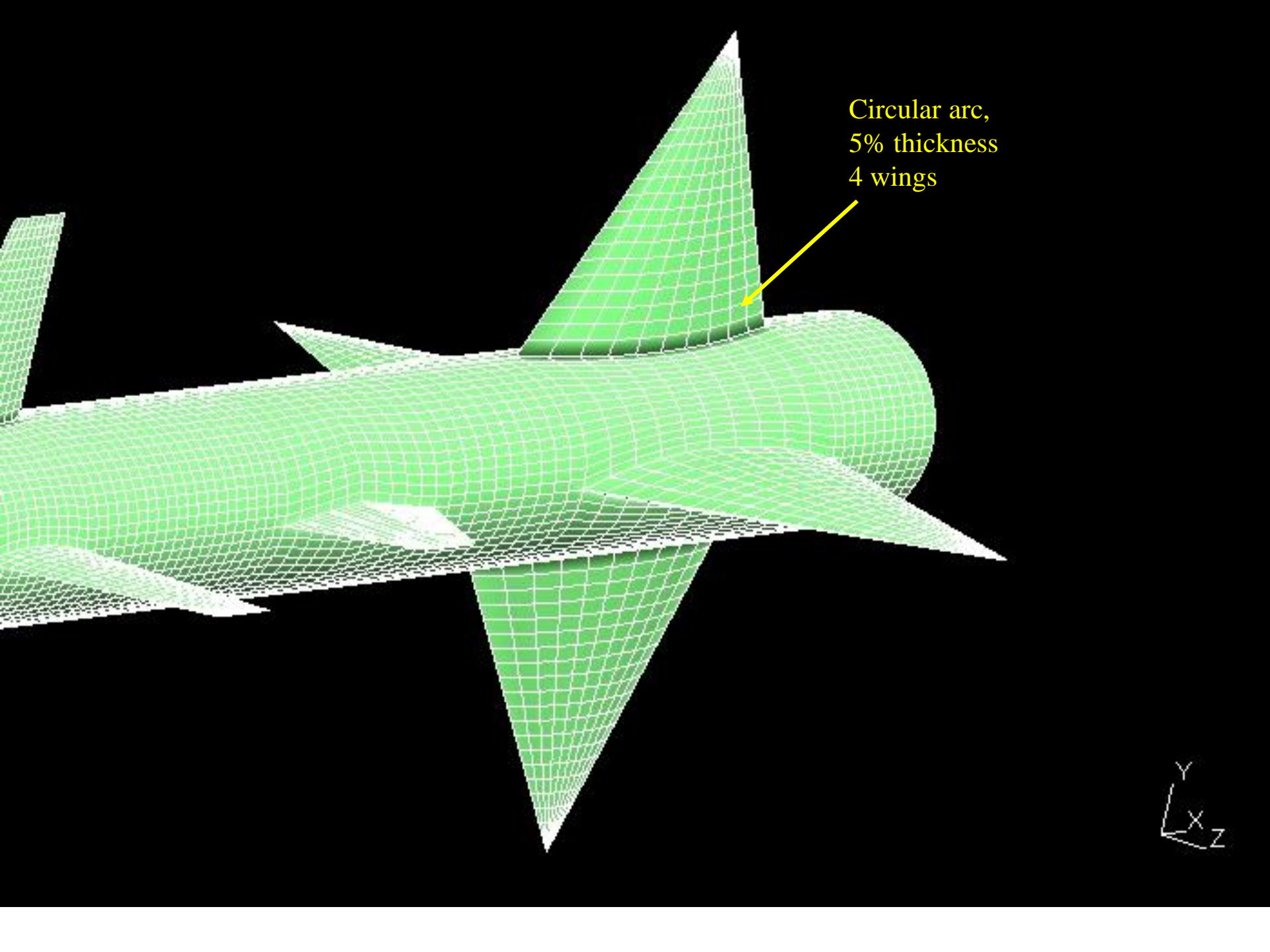
Circular arc,
5% thickness
4 wings



Front view of 3D model





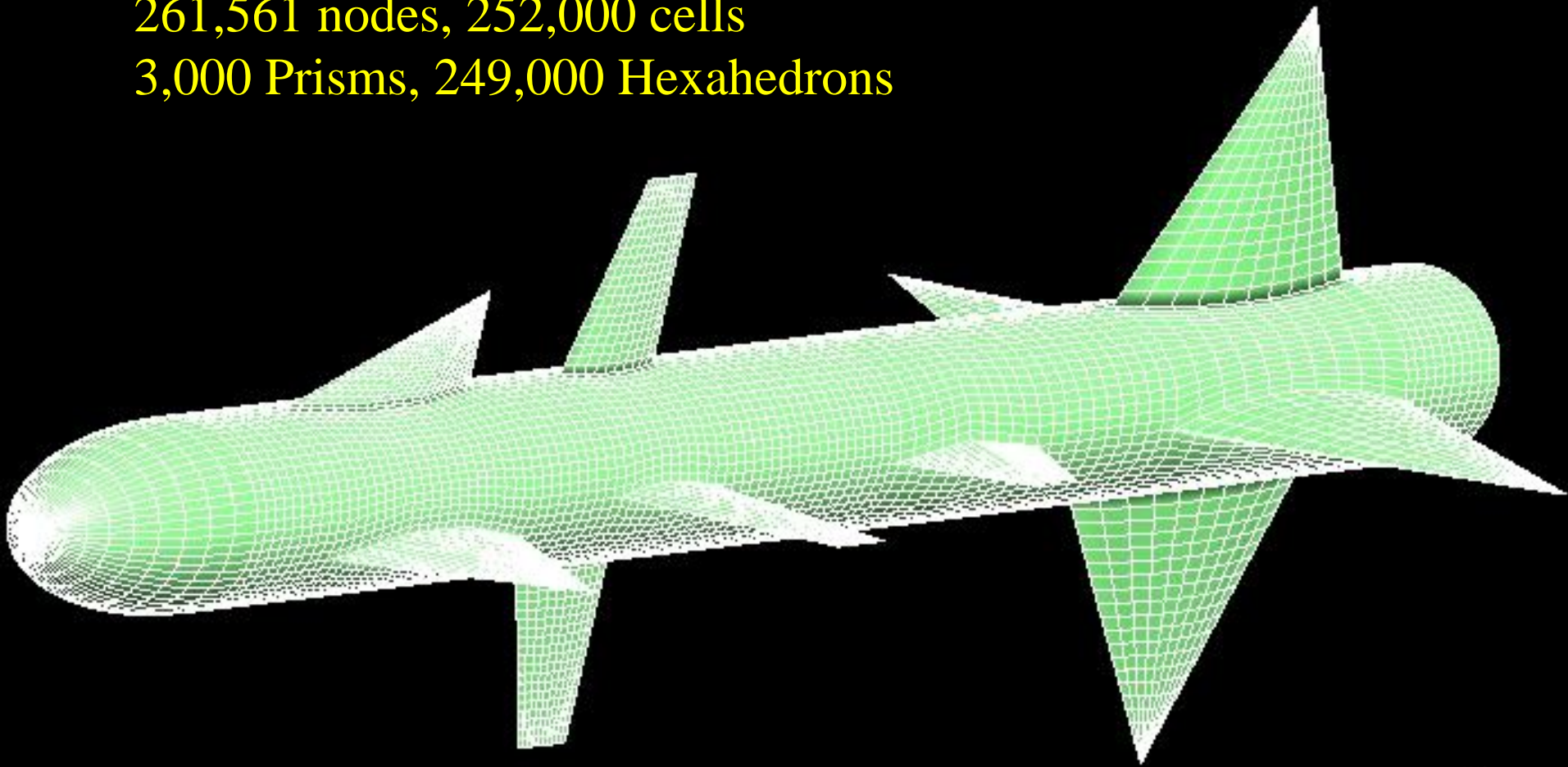


Circular arc,
5% thickness
4 wings

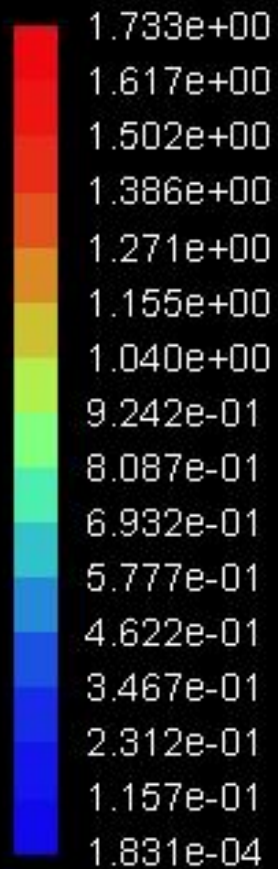
3D model

261,561 nodes, 252,000 cells

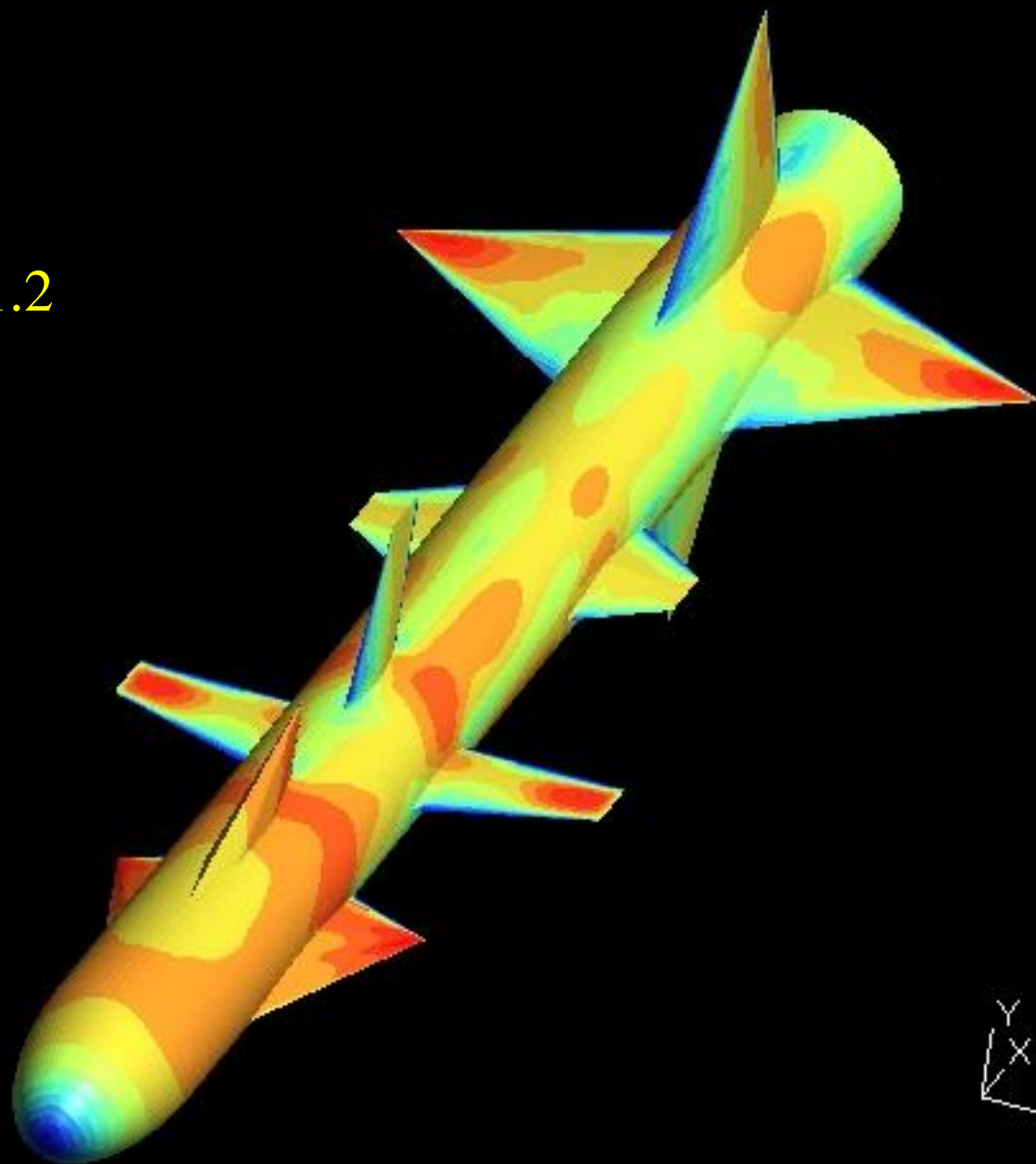
3,000 Prisms, 249,000 Hexahedrons



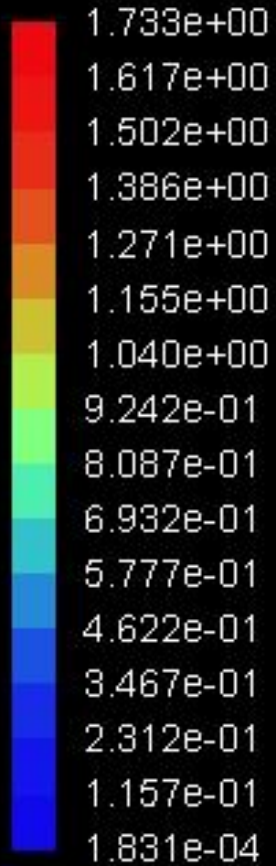
Mach = 1.2
 $\alpha = 10^\circ$



Mach number contour



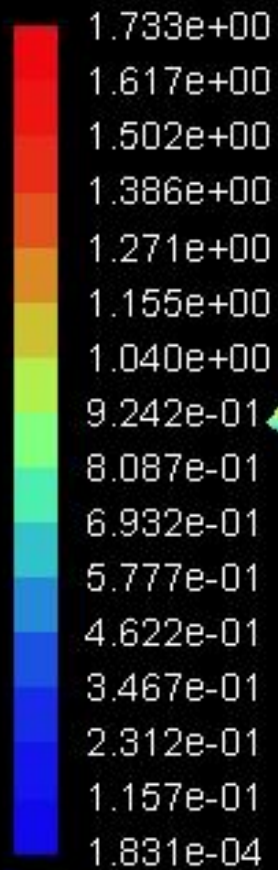
Mach = 1.2
 $\alpha = 10^\circ$



Mach number contour



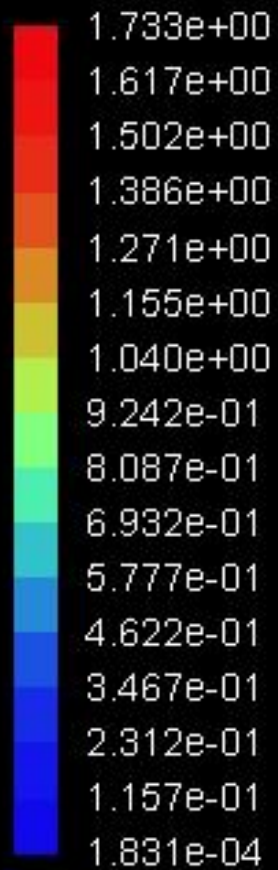
Mach = 1.2
 $\alpha = 10^\circ$



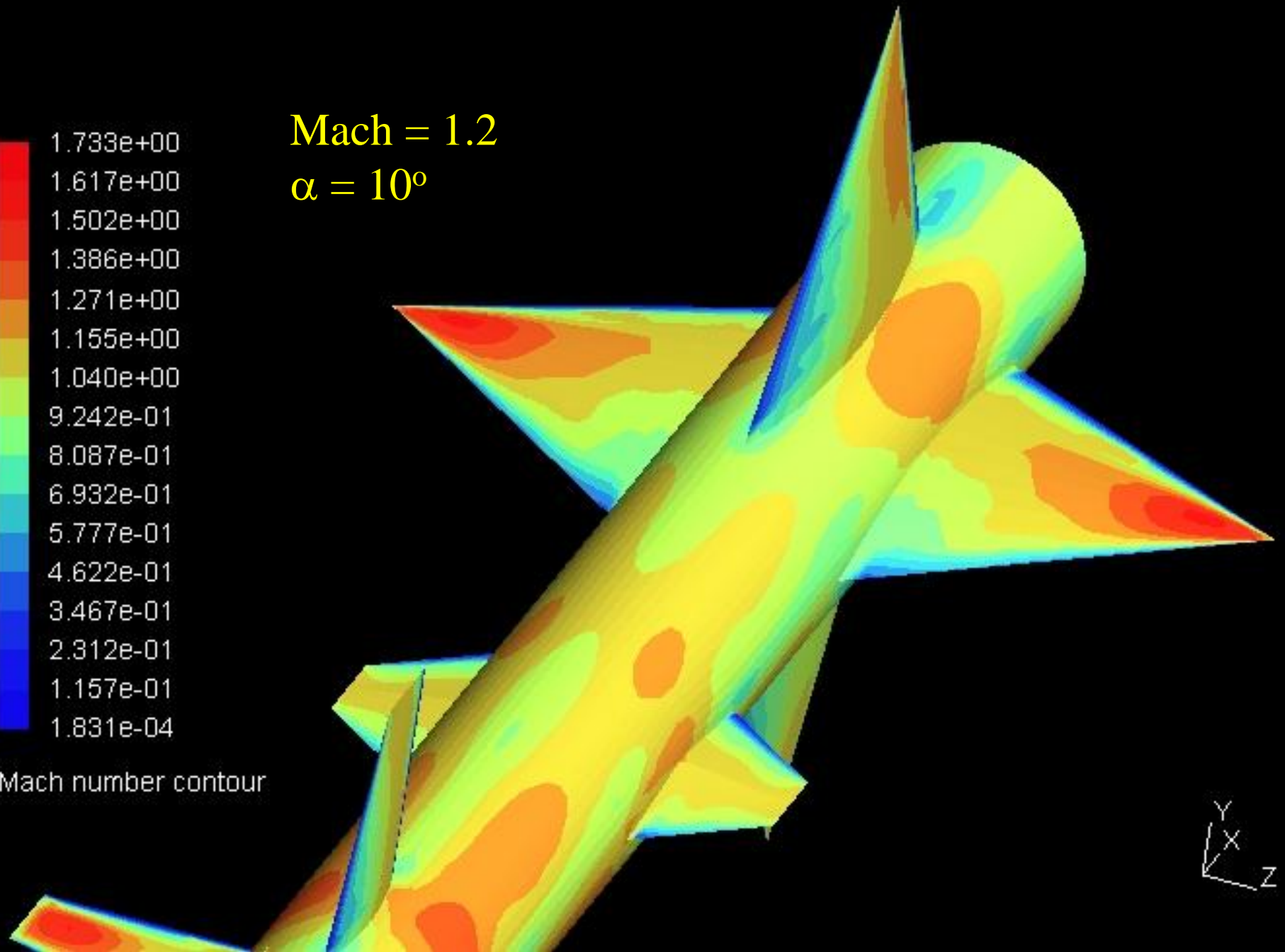
Mach number contour



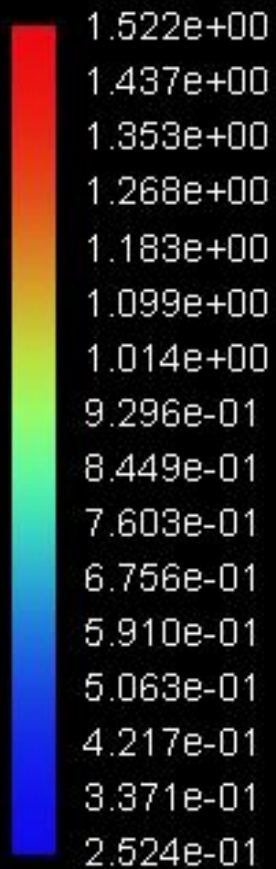
Mach = 1.2
 $\alpha = 10^\circ$



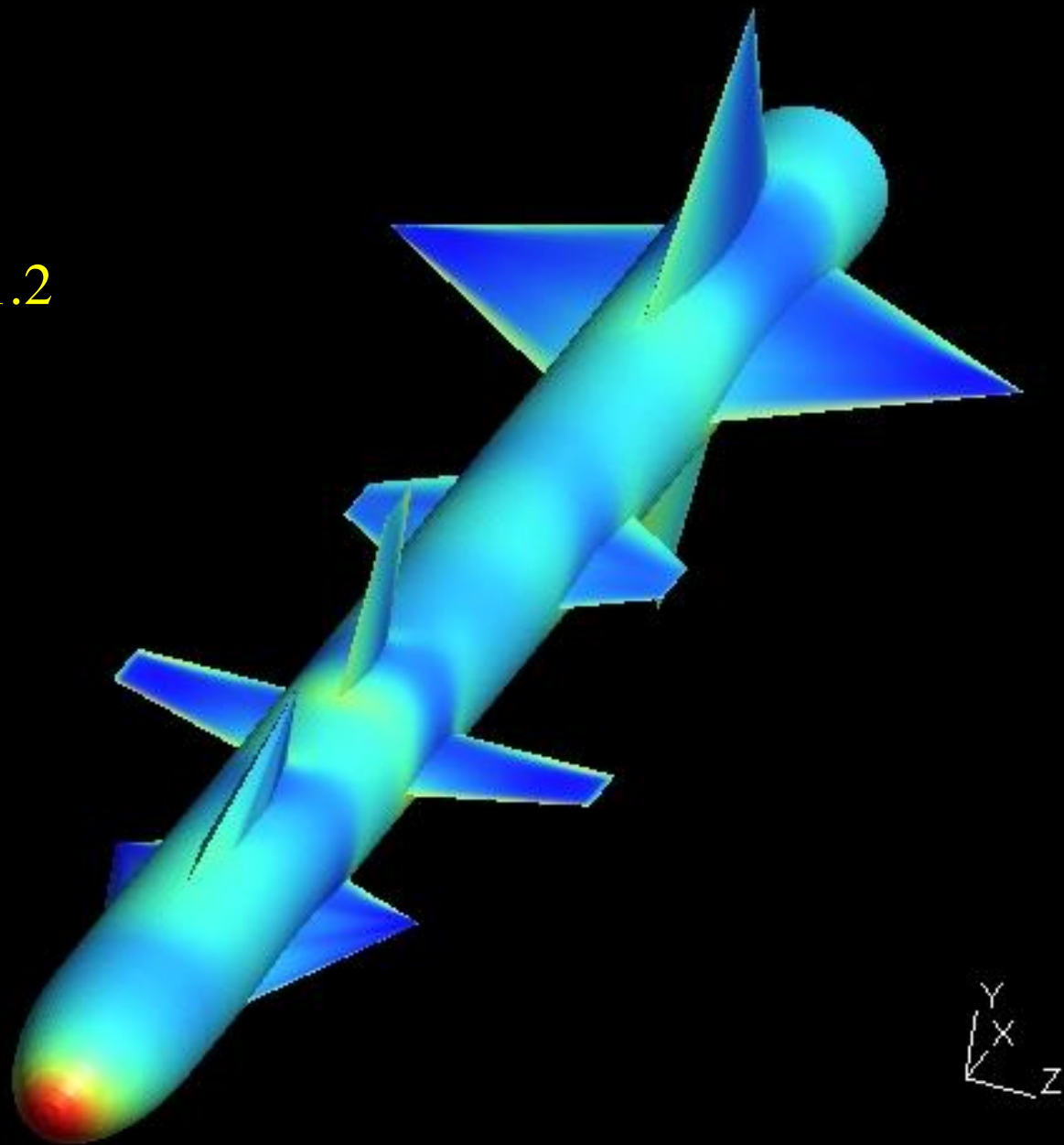
Mach number contour



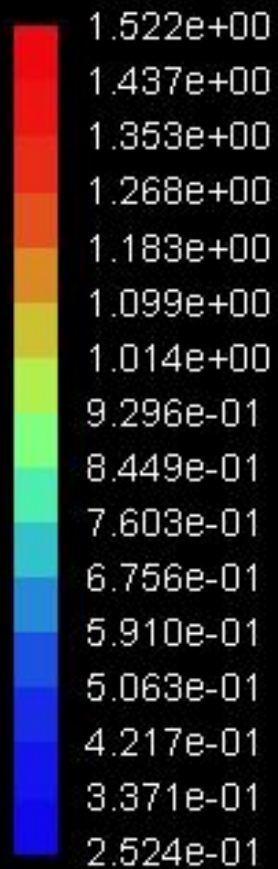
Mach = 1.2
 $\alpha = 10^\circ$



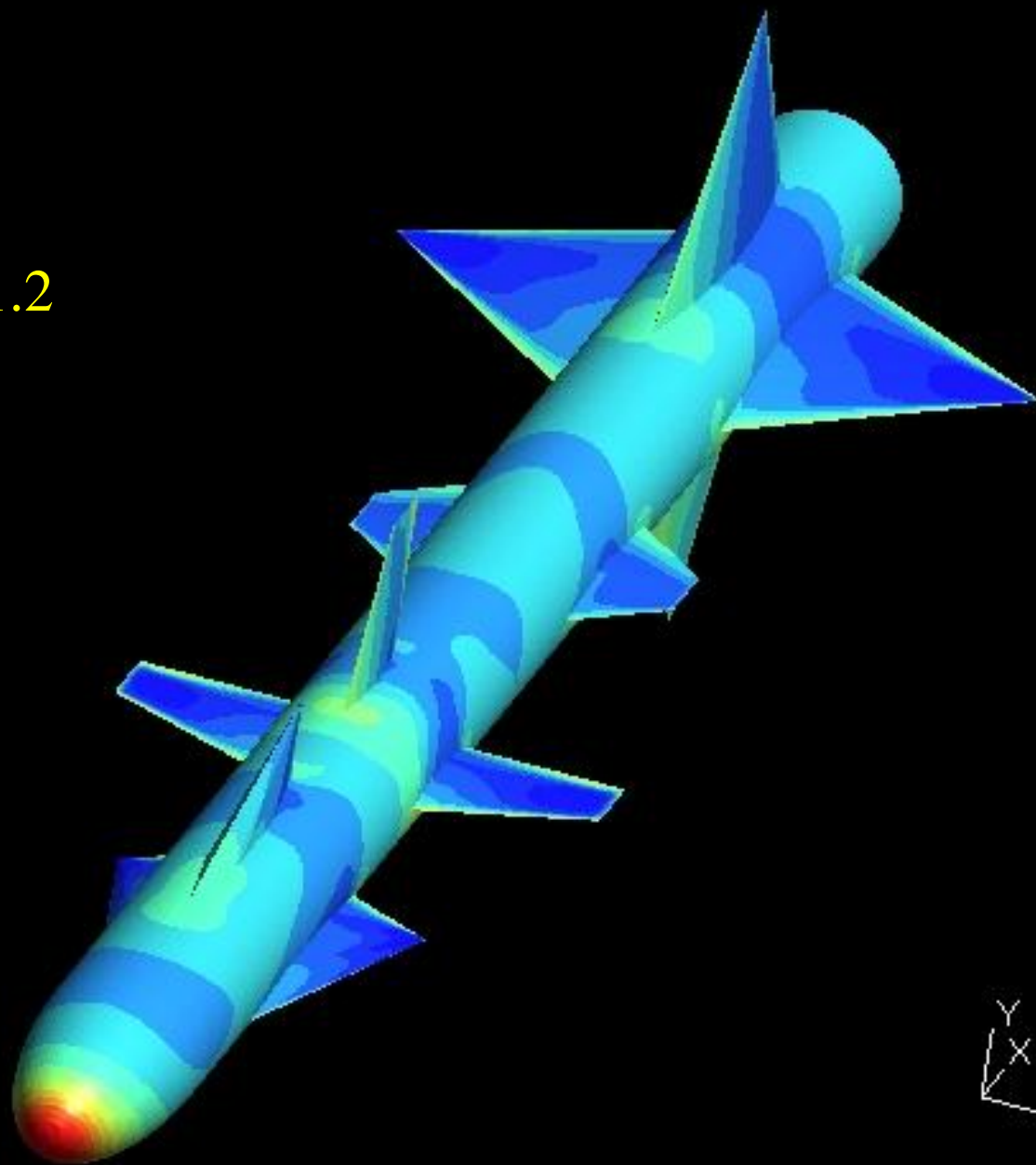
Pressure contour



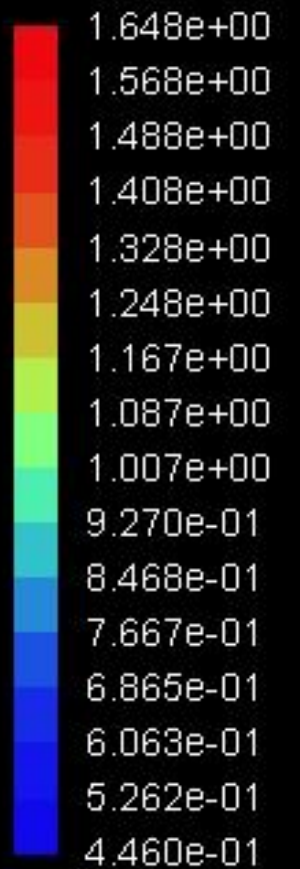
Mach = 1.2
 $\alpha = 10^\circ$



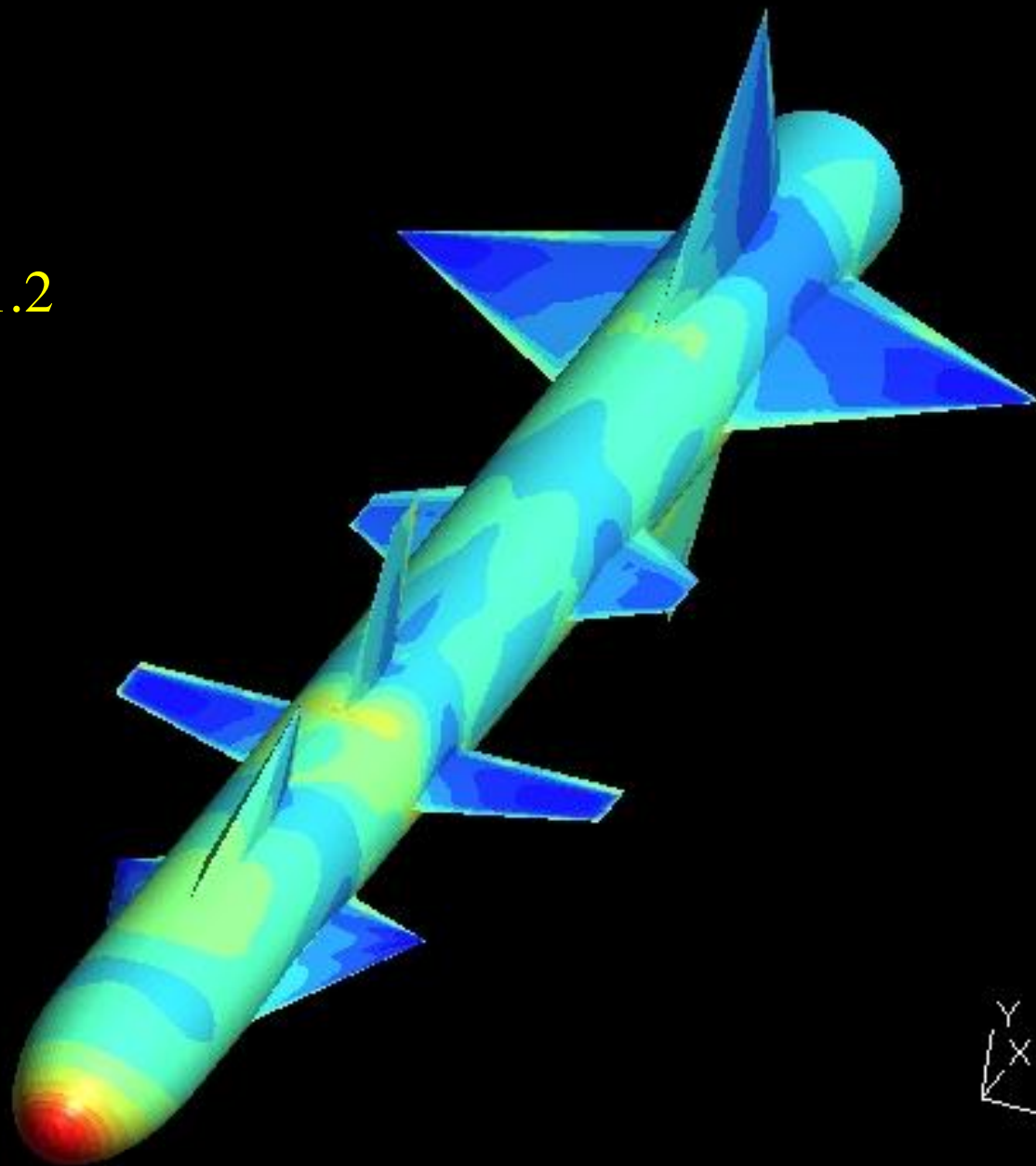
Pressure contour



Mach = 1.2
 $\alpha = 10^\circ$



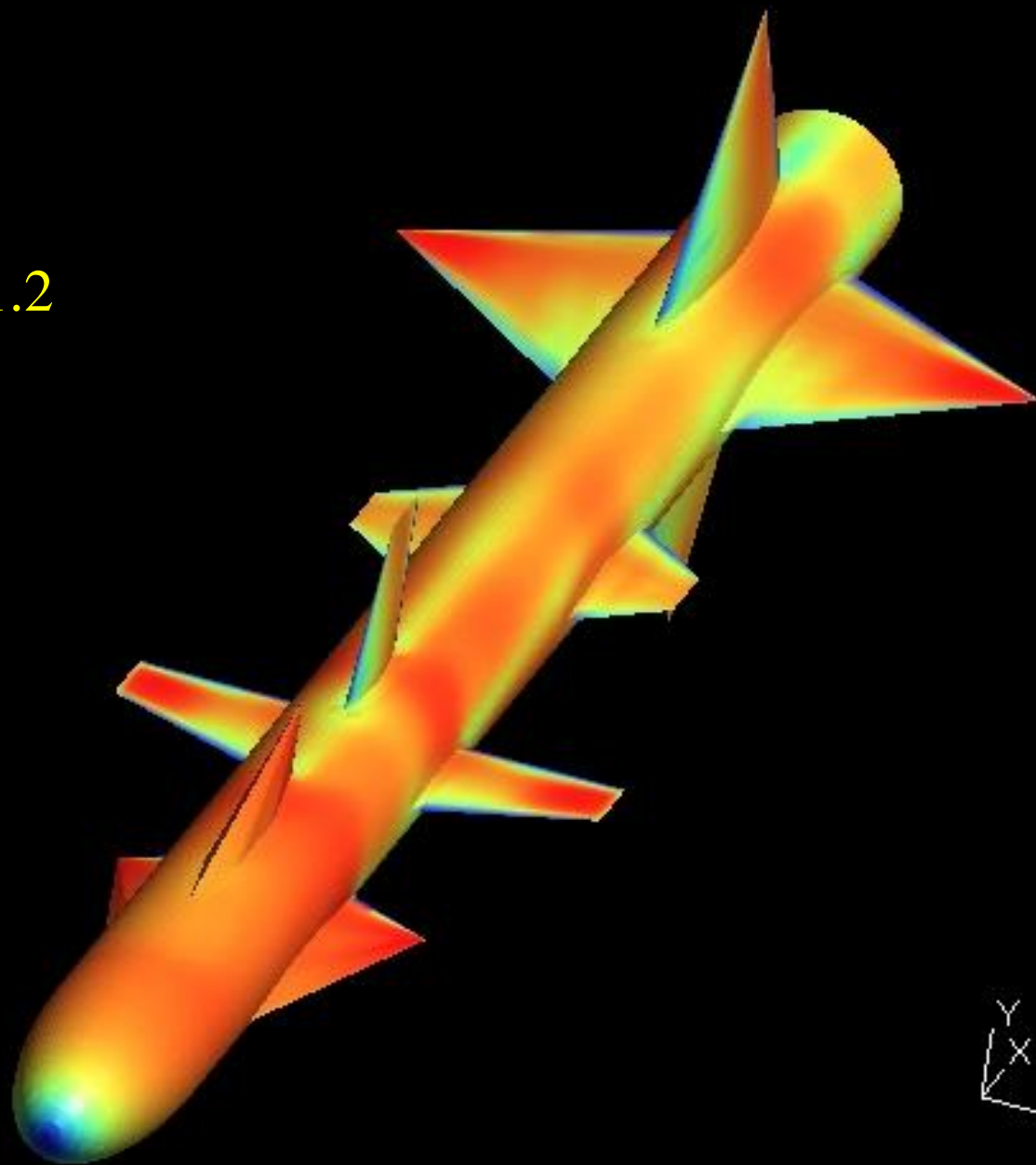
Density contour



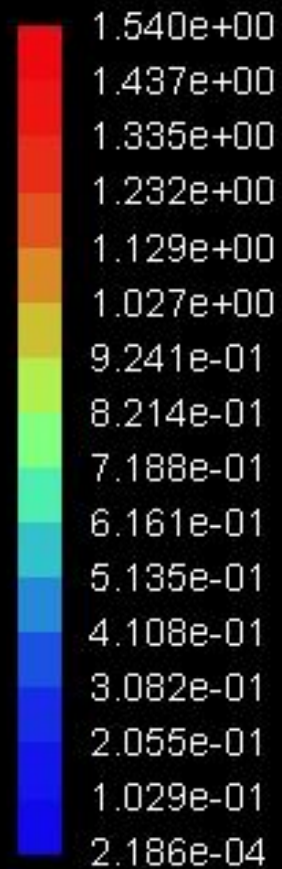
Mach = 1.2
 $\alpha = 10^\circ$



velocity magnitude plot



Mach = 1.2
 $\alpha = 10^\circ$



velocity magnitude plot

